

IN THE CLAIMS

1-8. (Previously cancelled).

9. (Currently Amended) A thermal barrier coating applied to a metallic component of a gas turbine engine comprising a top layer of a dense vertically cracked ceramic consisting of less than 1 to 6 weight % Ytria, 0-1 weight % Hafnia, and the balance Zirconia, and a bond coating adhering said ceramic layer to said metallic component, said bond coating comprising an oxidation-resistant alloy of MCrAlY with a thickness of about .003 to .025 inches, where M is iron, cobalt or nickel.

10. (Previously Added) A thermal barrier coating as recited in claim 9, wherein said dense vertically cracked ceramic layer is about 5-100 mils thick.

11. (Previously Added) A thermal barrier coating as recited in claim 9, wherein said bond coating consists of a diffusion aluminide or platinum aluminide that forms an oxidation-resistant intermetallic.

12. (Previously Added) The thermal barrier coating system as recited in claim 9, wherein said Zirconia is partially stabilized by Ytria.